

Weight and Balance Computation Sheet

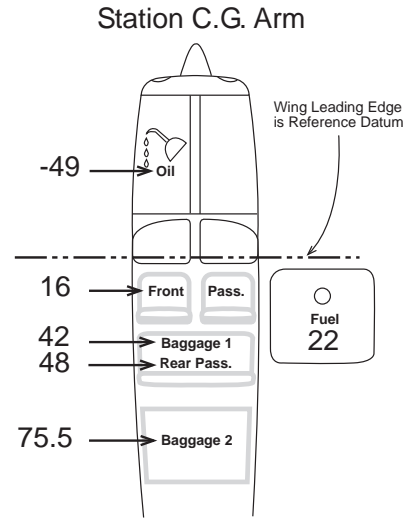
Stinson Voyager 108-1

	Weight x Arm = Moment		
Aircraft Empty			
Oil (8 qt. x 1.875 lb.)	15.0	-49.0	-735.00
Fuel (36 gal x 6 usable lb.)		16.0	
Front Seat Passenger		48.0	
Rear Seat Passenger		48.0	
Baggage 1 (Under Rear Seat)		42.0	
Baggage 2 (Compartment)		75.5	
Totals			
Total Moment / Total Weight = C.G. Inches			
<small>*Arm values obtained from Univair Publication Part# SSS</small>			

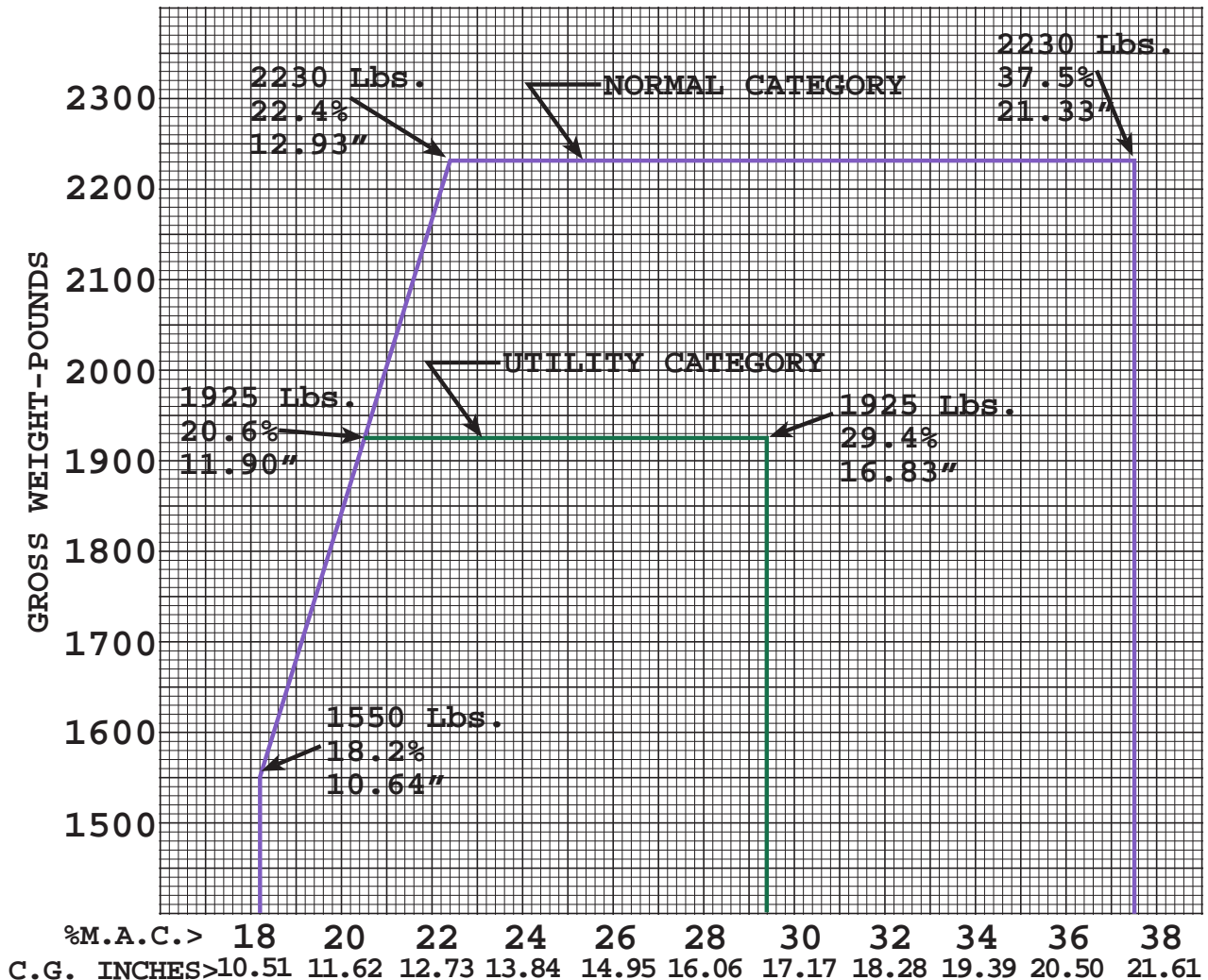
Weight Shift Formula

solve for X

$$\frac{\text{Weight Moved}}{\text{Weight of Plane}} = \frac{\text{Distance C.G. Moves}}{\text{Distance Between Arms}}$$



CENTER OF GRAVITY AND WEIGHT LIMITATIONS



To convert figures for %M.A.C. into C.G. Inches, divide %M.A.C. by 100, multiply by 55.50 and add .52 inches to this product [exa: ((%MAC/100)*55.50)+.52 = C.G. Inches]. To convert C.G. Inches to %M.A.C., subtract .52 from C.G. inches, divide by 55.50 and multiply by 100 [exa: ((C.G.-.52)/55.50)*100 = %M.A.C.].